

References

- ¹ Narayana, C. L. and Ramamoorthy, P., "Compressible Boundary-Layer Equations Solved by the Method of Parameter Differentiation," *AIAA Journal*, Vol. 10, No. 8, Aug. 1972, pp. 1085-1086.
- ² Cohen, C. B. and Reshotko, E., "Similar Solutions for the Compressible Laminar Boundary Layer with Heat Transfer and Pressure Gradient," TR-1293, 1956, NACA.

Reply by Authors to Tsung Y. Na

C. L. NARAYANA* AND P. RAMAMOORTHY†

National Aeronautical Laboratory, Bangalore, India

WE are thankful for Na's comments on our paper.¹ It is true that the expansion of the functions g and T in a three-term form as is done by Na, simplifies the procedure of solving the linear equations. However, the statement of Na that the value of the method of parameter differentiation lies in the elimination of iteration needs to be qualified for the following reasons. First, Eqs. (3-8) of his comments need iterations for solving if one uses predictor-corrector methods as has been done in Ref. 1. Secondly, the integrations of $\partial f/\partial \beta = g$ and $\partial S/\partial \beta = T$ [Eqs. (5) and (6) of Ref. 1] themselves need iterations if one uses trapezoidal rule as has been done in Ref. 1. Integration depending on iterative methods gives more accurate values for f and S and their derivatives which are to be used as coefficients in the equations for the components of g and T . To get a comparable accuracy by noniterative methods, one has to decrease the step length in β .

Reference

- ¹ Narayana, C. L. and Ramamoorthy, P., "Compressible Boundary-Layer Equations Solved by the Method of Parametric Differentiation," *AIAA Journal*, Vol. 10, No. 8, Aug. 1972, pp. 1085-1086.

Received June 25, 1973.

Index category: Boundary Layer and Convective Heat Transfer—Laminar.

* Scientist, Aerodynamics Division.

† Scientist, Aerodynamics Division. Member AIAA.

Comment on "Rates of Change of Flutter Mach Number and Flutter Frequency"

KUMAR G. BHATIA*

NASA Langley Research Center, Hampton, Va.

AND

CARL S. RUDISILL†

Clemson University, Clemson, S.C.

IN a recent technical note, Rao¹ published expressions for the rates of change of flutter Mach number and flutter frequency with respect to the structural design variables, and made

Received February 7, 1973.

Index category: Optimal Structural Design, Aeroelasticity.

* NRC-NASA Postdoctoral Resident Research Associate. Associate Member AIAA.

† Associate Professor, Department of Mechanical Engineering. Member AIAA.

reference to an earlier paper by Rudisill and Bhatia.² Rao has derived the expressions for the derivatives by separately considering the real and imaginary parts, and his procedure requires evaluation of the cofactors of the flutter determinant. It was shown in the paper by Rudisill and Bhatia that the two unknown derivatives which appear on differentiating the flutter equation can be determined by separating the real and imaginary parts of the differentiated equation, and their expressions require the eigenvectors only and not the cofactors. Therefore, the footnote in Rao's note referring to the paper by Rudisill and Bhatia should read "Their equation instead (not also) requires the eigenvectors of the flutter problem in order to compute $\partial V_F/\partial X_k$." Rao also states in the same footnote that they (Rudisill and Bhatia) have not used the expression to predict the flutter behavior at the perturbed design. In fact, Rudisill and Bhatia used the flutter velocity derivatives in their search scheme to compute the change in structural design variables necessary to obtain the desired flutter velocity. This is clearly stated in the text and illustrated in Fig. 3 of Ref. 2.

References

- ¹ Rao, S. S., "Rates of Change of Flutter Mach Number and Flutter Frequency," *AIAA Journal*, Vol. 10, No. 11, Nov. 1972, pp. 1525-1528.
- ² Rudisill, C. S. and Bhatia, K. G., "Optimization of Complex Structures to Satisfy Flutter Requirements," *AIAA Journal*, Vol. 9, No. 8, Aug. 1971, pp. 1487-1491.

Reply by Author to K. G. Bhatia and C. S. Rudisill

S. S. RAO*

Indian Institute of Technology, Kanpur, India

IT appears that Rudisill and Bhatia misunderstood the statements made by Rao in Ref. 1. The following points might clarify the matter.

1) In general, both eigenvalues and eigenvectors are required in order to compute the rates of change of eigenvectors of any real eigenvalue problem. Similarly, the expressions derived by Rudisill and Bhatia² for $\partial V_F/\partial X_k$ require both the eigenvalues and eigenvectors of the complex eigenvalue (flutter) problem. But the expression derived by Rao¹ requires only the eigenvalues of the flutter problem. The first statement made in the footnote of Ref. 1 has to be understood in this context.

2) In structural optimization, the rates of change of any behavior quantity can be used in two distinct ways. Firstly, the partial derivatives of the behavior quantity (like flutter speed) could be directly used in any first order optimization method to compute the necessary gradients. Secondly, the partial derivatives could be used in the analysis routine to predict approximately the behavior of the structure at any perturbed design. The second statement in the footnote of Ref. 1 has to be taken to mean that the derivatives $\partial V_F/\partial X_k$ have not been used in Ref. 2 for the latter purpose.

References

- ¹ Rao, S. S., "Rates of Change of Flutter Mach Number and Flutter Frequency," *AIAA Journal*, Vol. 10, No. 11, Nov. 1972, pp. 1526-1528.
- ² Rudisill, C. S. and Bhatia, K. G., "Optimization of Complex Structures to Satisfy Flutter Requirements," *AIAA Journal*, Vol. 9, No. 8, Aug. 1971, pp. 1487-1491.

Received April 11, 1973.

Index categories: Aeroelasticity and Hydroelasticity; Optimal Structural Design.

* Assistant Professor, Department of Mechanical Engineering.